

TITLE

ABSTRACT OF THE DISCLOSURE

An airfoil for a gas turbine engine includes a first sidewall and a second sidewall coupled together at a leading edge and a trailing edge, such that a cavity is defined therebetween. A plurality of rib walls extend at least partially between the first and second sidewalls, wherein the plurality of rib walls define at least one cooling circuit having at least three cooling chambers. At least one row of openings extend through at least one of the rib walls, wherein a first of the cooling chambers supplies cooling fluid to the cavity, and the remaining cooling chambers are coupled in flow communication with the first cooling chamber via the openings.